

Geometric Tracking Control of a Quadrotor UAV on $SE(3)$

Taeyoung Lee*, Melvin Leok†, and N. Harris McClamroch

Abstract—This paper provides new results for the tracking control of a quadrotor unmanned aerial vehicle (UAV). The UAV has four input degrees of freedom, namely the magnitudes of the four rotor thrusts, that are used to control the six translational and rotational degrees of freedom, and to achieve asymptotic tracking of four outputs, namely, three position variables for the vehicle center of mass and the direction of one vehicle body-fixed axis. A globally defined model of the quadrotor UAV rigid body dynamics is introduced as a basis for the analysis. A nonlinear tracking controller is developed on the special Euclidean group $SE(3)$ and it is shown to have desirable closed loop properties that are almost global. Several numerical examples, including an example in which the quadrotor recovers from being initially upside down, illustrate the versatility of the controller.

I. INTRODUCTION

A quadrotor unmanned aerial vehicle (UAV) consists of two pairs of counter-rotating rotors and propellers, located at the vertices of a square frame. It is capable of vertical take-off and landing (VTOL), but it does not require complex mechanical linkages, such as swash plates or teeter hinges, that commonly appear in typical helicopters. Due to its simple mechanical structure, it has been envisaged for various applications such as surveillance or mobile sensor networks as well as for educational purposes. There are several university-level projects [1], [2], [3], [4], and commercial products [5], [6], [7] related to the development and application of quadrotor UAVs.

Despite the substantial interest in quadrotor UAVs, little attention has been paid to constructing nonlinear control systems for them, particularly to designing nonlinear tracking controllers. Linear control systems such as proportional-derivative controllers or linear quadratic regulators are widely used to enhance the stability properties of an equilibrium [1], [3], [4], [8], [9]. A nonlinear controller is developed for the linearized dynamics of a quadrotor UAV with saturated positions in [10]. Backstepping and sliding mode techniques are applied in [11]. Since these controllers are based on Euler angles, they exhibit singularities when representing complex rotational maneuvers of a quadrotor UAV, thereby fundamentally restricting their ability to track nontrivial trajectories.

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Geometric control is concerned with the development of control systems for dynamic systems evolving on nonlinear manifolds that cannot be globally identified with Euclidean spaces [12], [13], [14]. By characterizing geometric properties of nonlinear manifolds intrinsically, geometric control techniques provide unique insights to control theory that cannot be obtained from dynamic models represented using local coordinates [15]. This approach has been applied to fully actuated rigid body dynamics on Lie groups to achieve almost global asymptotic stability [14], [16], [17], [18].

In this paper, we develop a geometric controller for a quadrotor UAV. The dynamics of a quadrotor UAV is expressed globally on the configuration manifold of the special Euclidean group $SE(3)$. We construct a tracking controller to follow prescribed trajectories for the center of mass and heading direction. It is shown that this controller exhibits almost global exponential attractiveness to the zero equilibrium of tracking errors. Since this is a coordinate-free control approach, it completely avoids singularities and complexities that arise when using local coordinates.

Compared to other geometric control approaches for rigid body dynamics, this is distinct in the sense that it controls an underactuated quadrotor UAV to stabilize six translational and rotational degrees of freedom using four thrust inputs, while asymptotically tracking four outputs consisting of its position and heading direction. We demonstrate that this controller is particularly useful for complex, acrobatic maneuvers of a quadrotor UAV, such as recovering from being initially upside down.

II. QUADROTOR DYNAMICS MODEL

Consider a quadrotor vehicle model illustrated in Fig. 1. This is a system of four identical rotors and propellers located at the vertices of a square, which generate a thrust and torque normal to the plane of this square.

We choose an inertial reference frame $\{\vec{i}_1, \vec{i}_2, \vec{i}_3\}$ and a body-fixed frame $\{\vec{b}_1, \vec{b}_2, \vec{b}_3\}$. The origin of the body-fixed frame is located at the center of mass of this vehicle. The first and the second axes of the body fixed frame, \vec{b}_1, \vec{b}_2 , lie in the plane defined by the centers of the four rotors, as illustrated in Fig. 1. The third body fixed axis \vec{b}_3 is normal to this plane, and points downward, opposite to the direction of the total thrust. Define

- $m \in \mathbb{R}$ the total mass
- $J \in \mathbb{R}^{3 \times 3}$ the inertia matrix with respect to the body-fixed frame
- $R \in SO(3)$ the rotation matrix from the body-fixed frame to the inertial frame
- $\Omega \in \mathbb{R}^3$ the angular velocity in the body-fixed frame

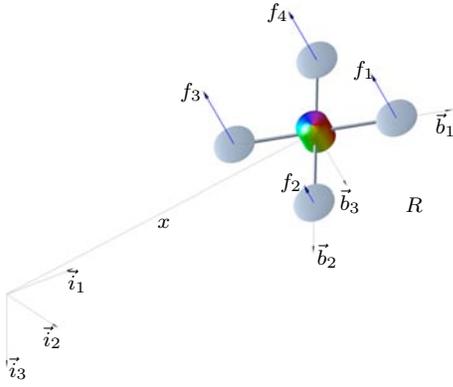


Fig. 1. Quadrotor model

$x \in \mathbb{R}^3$	the location of the center of mass in the inertial frame
$v \in \mathbb{R}^3$	the velocity of the center of mass in the inertial frame
$d \in \mathbb{R}$	the distance from the center of mass to the center of each rotor in the \vec{b}_1, \vec{b}_2 plane
$f_i \in \mathbb{R}$	the thrust generated by the i -th propeller along the $-\vec{b}_3$ axis
$\tau_i \in \mathbb{R}$	the torque generated by the i -th propeller about the \vec{b}_3 axis
$f \in \mathbb{R}$	the total thrust, i.e., $f = \sum_{i=1}^4 f_i$
$M \in \mathbb{R}^3$	the total moment in the body-fixed frame

The configuration of this quadrotor UAV is defined by the location of the center of mass and the attitude with respect to the inertial frame. Therefore, the configuration manifold is the special Euclidean group $SE(3)$, which is the semidirect product of \mathbb{R}^3 and the special orthogonal group $SO(3) = \{R \in \mathbb{R}^{3 \times 3} \mid R^T R = I, \det R = 1\}$.

We assume that the thrust of each propeller is directly controlled, i.e., we do not consider the dynamics of rotors and propellers, and the direction of the thrust is normal to the quadrotor plane. The total thrust is $f = \sum_{i=1}^4 f_i$, which acts along the direction of $-\vec{b}_3$. According to the definition of the rotation matrix $R \in SO(3)$, the direction of the i -th body fixed axis \vec{b}_i is given by Re_i in the inertial frame, where $e_1 = [1; 0; 0], e_2 = [0; 1; 0], e_3 = [0; 0; 1] \in \mathbb{R}^3$. Therefore, the total thrust is given by $-fRe_3 \in \mathbb{R}^3$ in the inertial frame.

We also assume that the torque generated by each propeller is directly proportional to its thrust. Since it is assumed that the first and the third propellers rotate clockwise, and the second and the fourth propellers rotate counterclockwise, when they are generating a positive thrust f_i along $-\vec{b}_3$, the torque generated by the i -th propeller can be written as $\tau_i = (-1)^i c_{\tau f} f_i$ for a fixed constant $c_{\tau f}$. All of these assumptions are common [19], [4], and the presented control system can readily be extended to include linear rotor dynamics studied in [11]. Under these assumptions, the total thrust f and the total moment M can be written as

$$\begin{bmatrix} f \\ M_1 \\ M_2 \\ M_3 \end{bmatrix} = \begin{bmatrix} 1 & 1 & 1 & 1 \\ 0 & -d & 0 & d \\ d & 0 & -d & 0 \\ -c_{\tau f} & c_{\tau f} & -c_{\tau f} & c_{\tau f} \end{bmatrix} \begin{bmatrix} f_1 \\ f_2 \\ f_3 \\ f_4 \end{bmatrix}. \quad (1)$$

The determinant of the above 4×4 matrix is $8c_{\tau f}d^2$, so it is invertible when $d \neq 0$ and $c_{\tau f} \neq 0$. Therefore, for a given f, M , the thrust of each rotor f_i can be obtained from (1). Using this equation, the total thrust $f \in \mathbb{R}$ and the moment $M \in \mathbb{R}^3$ are considered as control inputs in this paper.

The equations of motion of this quadrotor UAV can be written as

$$\dot{x} = v, \quad (2)$$

$$m\dot{v} = mge_3 - fRe_3, \quad (3)$$

$$\dot{R} = R\hat{\Omega}, \quad (4)$$

$$J\dot{\Omega} + \Omega \times J\Omega = M, \quad (5)$$

where the *hat map* $\hat{\cdot} : \mathbb{R}^3 \rightarrow \mathfrak{so}(3)$ is defined by the condition that $\hat{x}y = x \times y$ for all $x, y \in \mathbb{R}^3$.

III. GEOMETRIC TRACKING CONTROL ON $SE(3)$

We develop a controller to follow a prescribed trajectory of the location of the center of mass, $x_d(t)$, and the desired direction of the first body-fixed axis, $\vec{b}_{1_d}(t)$.

The overall controller structure is as follows. The translational dynamics of a quadrotor UAV is controlled by the total thrust $-fRe_3$, where the magnitude of the total thrust f is directly controlled, and the direction of the total thrust $-Re_3$ is along the third body-fixed axis $-\vec{b}_3$. For a given translational command $x_d(t)$, we select the total thrust f , and the desired direction of the third body-fixed axis \vec{b}_{3_d} to stabilize the translational dynamics.

Once the desired direction of the third body-fixed frame \vec{b}_{3_d} is chosen, there is one remaining degree of freedom in selecting the desired attitude, $R_d \in SO(3)$. This corresponds to the heading direction of the quadrotor UAV in the plane normal to \vec{b}_{3_d} , and it is determined by the desired direction of the first body-fixed axis, $\vec{b}_{1_d}(t)$. More explicitly, we assume that \vec{b}_{1_d} is not parallel to \vec{b}_{3_d} , and we project it onto the plane normal to \vec{b}_{3_d} to obtain the complete desired attitude: $R_d = [\vec{b}_{2_d} \times \vec{b}_{3_d}, \vec{b}_{2_d}, \vec{b}_{3_d}] \in SO(3)$, where $\vec{b}_{2_d} = (\vec{b}_{3_d} \times \vec{b}_{1_d}) / \|\vec{b}_{3_d} \times \vec{b}_{1_d}\|$. The control moment M is selected to follow this desired attitude.

In short, four-dimensional control inputs are designed to follow a three-dimensional translational command, and a one-dimensional heading direction, i.e., we guarantee that $x(t) \rightarrow x_d(t)$ and $\text{Proj}[\vec{b}_1(t)] \rightarrow \text{Proj}[\vec{b}_{1_d}(t)]$ as $t \rightarrow \infty$, where $\text{Proj}[\cdot]$ denotes the normalized projection onto the plane orthogonal to \vec{b}_{3_d} . The overall controller structure and the definition of the heading direction are illustrated in Fig. 2 and 3.

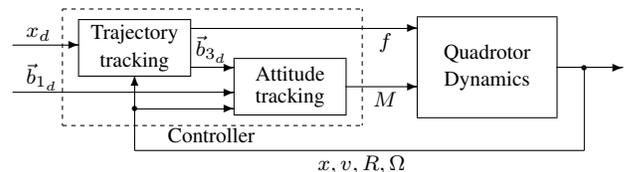


Fig. 2. Controller structure

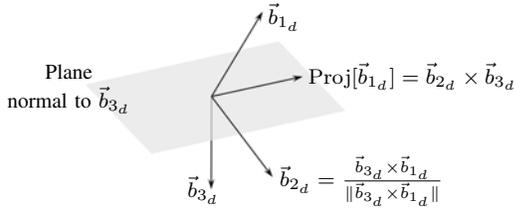


Fig. 3. Definition of the desired heading direction

We develop this controller directly on the nonlinear configuration Lie group and thereby avoid any singularities and complexities that arise in local coordinates. As a result, we are able to achieve almost global exponential attractiveness to the zero equilibrium of tracking errors. Due to page limit constraint, all of the proofs are relegated to [20].

A. Tracking Errors

We define the tracking errors for x, v, R, Ω as follows. The tracking errors for the position and the velocity are given by:

$$e_x = x - x_d, \quad (6)$$

$$e_v = v - v_d. \quad (7)$$

The attitude and angular velocity tracking error should be carefully chosen as they evolve on the tangent bundle of the nonlinear space $\text{SO}(3)$. The error function on $\text{SO}(3)$ is chosen to be

$$\Psi(R, R_d) = \frac{1}{2} \text{tr}[I - R_d^T R]. \quad (8)$$

This is locally positive-definite about $R = R_d$ within the region where the rotation angle between R and R_d is less than 180° [14]. This set can be represented by the sublevel set of Ψ where $\Psi < 2$, namely $L_2 = \{R_d, R \in \text{SO}(3) \mid \Psi(R, R_d) < 2\}$, which almost covers $\text{SO}(3)$. When the variation of the rotation matrix is expressed as $\delta R = R\hat{\eta}$ for $\eta \in \mathbb{R}^3$, the derivative of the error function is given by

$$\begin{aligned} \mathbf{D}_R \Psi(R, R_d) \cdot R\hat{\eta} &= -\frac{1}{2} \text{tr}[R_d^T R\hat{\eta}] \\ &= \frac{1}{2} (R_d^T R - R^T R_d)^\vee \cdot \eta, \end{aligned} \quad (9)$$

where the *vee map* $^\vee : \mathfrak{so}(3) \rightarrow \mathbb{R}^3$ is the inverse of the hat map. We used the fact that $-\frac{1}{2} \text{tr}[\hat{x}\hat{y}] = x^T y$ for any $x, y \in \mathbb{R}^3$. From this, the attitude tracking error e_R is chosen to be

$$e_R = \frac{1}{2} (R_d^T R - R^T R_d)^\vee. \quad (10)$$

The tangent vectors $\dot{R} \in \mathbb{T}_R \text{SO}(3)$ and $\dot{R}_d \in \mathbb{T}_{R_d} \text{SO}(3)$ cannot be directly compared since they lie in different tangent spaces. We transform \dot{R}_d into a vector in $\mathbb{T}_R \text{SO}(3)$, and we compare it with \dot{R} as follows:

$$\dot{R} - \dot{R}_d (R_d^T R) = R\hat{\Omega} - R_d \hat{\Omega}_d R_d^T R = R(\Omega - R^T R_d \Omega_d)^\wedge.$$

We choose the tracking error for the angular velocity as follows:

$$e_\Omega = \Omega - R^T R_d \Omega_d. \quad (11)$$

We can show that e_Ω is the angular velocity of the rotation matrix $R_d^T R$, represented in the body-fixed frame, since $\frac{d}{dt} (R_d^T R) = (R_d^T R) \hat{e}_\Omega$.

B. Tracking Controller

For given smooth tracking commands $x_d(t), \vec{b}_{1_d}(t)$, and some positive constants k_x, k_v, k_R, k_Ω , we define

$$\vec{b}_{3_d} = -\frac{-k_x e_x - k_v e_v - m g e_3 + m \ddot{x}_d}{\| -k_x e_x - k_v e_v - m g e_3 + m \ddot{x}_d \|}, \quad (12)$$

where we assume that

$$\| -k_x e_x - k_v e_v - m g e_3 + m \ddot{x}_d \| \neq 0. \quad (13)$$

We also assume that \vec{b}_{1_d} is not parallel to \vec{b}_{3_d} . The desired attitude is given by $R_d = [\vec{b}_{2_d} \times \vec{b}_{3_d}, \vec{b}_{2_d}, \vec{b}_{3_d}] \in \text{SO}(3)$, where $\vec{b}_{2_d} = (\vec{b}_{3_d} \times \vec{b}_{1_d}) / \|\vec{b}_{3_d} \times \vec{b}_{1_d}\|$. The desired trajectory satisfies

$$\| -m g e_3 + m \ddot{x}_d \| < B \quad (14)$$

for a given positive constant B . The control inputs f, M are chosen as follows:

$$f = -(-k_x e_x - k_v e_v - m g e_3 + m \ddot{x}_d) \cdot R e_3, \quad (15)$$

$$\begin{aligned} M &= -k_R e_R - k_\Omega e_\Omega + \Omega \times J \Omega \\ &\quad - J(\hat{\Omega} R^T R_d \Omega_d - R^T R_d \hat{\Omega}_d). \end{aligned} \quad (16)$$

The control moment M given in (16) corresponds to a tracking controller on $\text{SO}(3)$. For the attitude dynamics of a rigid body described by (4), (5), this controller exponentially stabilizes the zero equilibrium of the attitude tracking errors.

Similarly, the expression in the parentheses in (15) corresponds to a tracking controller for the translational dynamics on \mathbb{R}^3 . The total thrust f and the desired direction \vec{b}_{3_d} of the third body-fixed axis are chosen such that if there is no attitude tracking error, the thrust vector term $-f R e_3$ in the translational dynamics of (3) becomes this tracking controller in \mathbb{R}^3 . Therefore, the translational tracking error will converge to zero provided that the attitude tracking error is identically zero.

Certainly, the attitude tracking error may not be zero at any instant. As the attitude tracking error increases, the direction of the control input term $f R e_3$ of the translational dynamics deviates from the desired direction $R_d e_3$. This may cause instability in the complete dynamics. In (15), we carefully design the magnitude of the total thrust f so that it is reduced when there is a large attitude tracking error. The expression of f includes the dot product of the desired third body-fixed axis $\vec{b}_{3_d} = R_d e_3$ and the current third body-fixed axis $\vec{b}_3 = R e_3$. Therefore, the magnitude of f is reduced when the angle between those two axes becomes larger. These effects are carefully analyzed in the stability proof for the complete dynamics in [20].

In short, this control system is designed so that the position tracking error converges to zero when there is no attitude tracking error, and it is properly adjusted for non-zero attitude tracking errors to achieve asymptotic stability of the complete dynamics.

C. Exponential Asymptotic Stability

We first show exponential stability of the attitude dynamics in the sublevel set $L_2 = \{R_d, R \in \text{SO}(3) \mid \Psi(R, R_d) < 2\}$, and based on this result, we show exponential stability

of the complete dynamics in the smaller sublevel set $L_1 = \{R_d, R \in \text{SO}(3) \mid \Psi(R, R_d) < 1\}$

Proposition 1: (Exponential Stability of Attitude Dynamics) Consider the control moment M defined in (16) for any positive constants k_R, k_Ω . Suppose that the initial condition satisfies

$$\Psi(R(0), R_d(0)) < 2, \quad (17)$$

$$\|e_{\Omega}(0)\|^2 < \frac{2}{\lambda_{\max}(J)} k_R (2 - \Psi(R(0), R_d(0))), \quad (18)$$

where $\lambda_{\min}(J)$ denotes the minimum eigenvalue of the inertia matrix J . Then, the zero equilibrium of the attitude tracking error e_R, e_Ω is exponentially stable. Furthermore, there exist constants $\alpha_2, \beta_2 > 0$ such that

$$\Psi(R(t), R_d(t)) \leq \min\{2, \alpha_2 e^{-\beta_2 t}\}. \quad (19)$$

Proof: See [20]. ■

In this proposition, (17), (18) represent a region of attraction for the attitude dynamics. This requires that the initial attitude error should be less than 180° . Therefore, the region of attraction for the attitude almost covers $\text{SO}(3)$, and the region of attraction for the angular velocity can be increased by choosing a larger controller gain k_R in (18).

Since the direction of the total thrust is the third body-fixed axis, the stability of the translational dynamics depends on the attitude tracking error. More precisely, the position tracking performance is affected by the difference between $\vec{b}_3 = R e_3$ and $\vec{b}_{3_d} = R_d e_3$. In the proceeding stability analysis, it turns out that for the stability of the complete translational and rotational dynamics, the attitude error function Ψ should be less than 1, which states that the initial attitude error should be less than 90° . For the stability of the complete system, we restrict the initial attitude error to obtain the following proposition.

Proposition 2: (Exponential Stability of the Complete Dynamics) Consider the control force f and moment M defined in (15), (16). Suppose that the initial condition satisfies

$$\Psi(R(0), R_d(0)) \leq \psi_1 < 1 \quad (20)$$

for a fixed constant ψ_1 . Define $W_1, W_{12}, W_2 \in \mathbb{R}^{2 \times 2}$ to be

$$W_1 = \begin{bmatrix} \frac{c_1 k_x}{m} & -\frac{c_1 k_v}{2m} (1 + \alpha) \\ -\frac{c_1 k_x}{2m} (1 + \alpha) & k_v (1 - \alpha) - c_1 \end{bmatrix}, \quad (21)$$

$$W_{12} = \begin{bmatrix} k_x e_{v_{\max}} + \frac{c_1}{m} B & 0 \\ B & 0 \end{bmatrix}, \quad (22)$$

$$W_2 = \begin{bmatrix} \frac{c_2 k_R}{\lambda_{\max}(J)} & -\frac{c_2 k_\Omega}{2\lambda_{\min}(J)} \\ -\frac{c_2 k_\Omega}{2\lambda_{\min}(J)} & k_\Omega - c_2 \end{bmatrix}, \quad (23)$$

where $\alpha = \sqrt{\psi_1(2 - \psi_1)}$, $e_{v_{\max}} = \max\{\|e_v(0)\|, \frac{B}{k_v(1 - \alpha)}\}$, $c_1, c_2 \in \mathbb{R}$. For any positive constants k_x, k_v , we choose positive constants c_1, c_2, k_R, k_Ω such that

$$c_1 < \min\left\{k_v(1 - \alpha), \frac{4mk_x k_v(1 - \alpha)}{k_v^2(1 + \alpha)^2 + 4mk_x}, \sqrt{k_x m}\right\}, \quad (24)$$

$$c_2 < \min\left\{k_\Omega, \frac{4k_\Omega k_R \lambda_{\min}(J)^2}{k_\Omega^2 \lambda_{\max}(J) + 4k_R \lambda_{\min}(J)^2}, \sqrt{k_R \lambda_{\min}(J)}, \sqrt{\frac{2}{2 - \psi_1} k_R \lambda_{\max}(J)}\right\}, \quad (25)$$

$$\lambda_{\min}(W_2) > \frac{4\|W_{12}\|^2}{\lambda_{\min}(W_1)}. \quad (26)$$

Then, the zero equilibrium of the tracking errors of the complete system is exponentially stable. The region of attraction is characterized by (20) and

$$\|e_\Omega(0)\|^2 < \frac{2}{\lambda_{\max}(J)} k_R (1 - \Psi(R(0), R_d(0))). \quad (27)$$

Proof: See [20]. ■

D. Almost Global Exponential Attractiveness

Proposition 2 requires that the initial attitude error is less than 90° to achieve exponential stability of the complete dynamics. Suppose that this is not satisfied, i.e. $1 \leq \Psi(R(0), R_d(0)) < 2$. From Proposition 1, we are guaranteed that the attitude error function Ψ exponentially decreases, and therefore, it enters the region of attraction of Proposition 2 in a finite time. Therefore, by combining the results of Proposition 1 and 2, we can show almost global exponential attractiveness.

Definition 1: (Exponential Attractiveness [21]) An equilibrium point $z = 0$ of a dynamic systems is *exponentially attractive* if, for some $\delta > 0$, there exists a constant $\alpha(\delta) > 0$ and $\beta > 0$ such that $\|z(0)\| < \delta \Rightarrow \|z(t)\| \leq \alpha(\delta) e^{-\beta t}$ for any $t > 0$.

This should be distinguished from the stronger notion of exponential stability, in which the constant $\alpha(\delta)$ in the above bound is replaced by $\alpha(\delta) \|z(0)\|$.

Proposition 3: (Almost Global Exponential Attractiveness of the Complete Dynamics) Consider a control system designed according to Proposition 2. Suppose that the initial condition satisfies

$$1 \leq \Psi(R(0), R_d(0)) < 2, \quad (28)$$

$$\|e_\Omega(0)\|^2 < \frac{2}{\lambda_{\max}(J)} k_R (2 - \Psi(R(0), R_d(0))). \quad (29)$$

Then, the zero equilibrium of the tracking errors of the complete dynamics is exponentially attractive.

Proof: See [20]. ■

Since the region of attraction given by (28) for the attitude almost covers $\text{SO}(3)$, this is referred to as almost global exponential attractiveness in this paper. The region of attraction for the angular velocity can be expanded by choosing a large gain k_R in (29).

E. Properties and Extensions

One of the unique properties of the presented controller is that it is directly developed on $\text{SE}(3)$ using rotation matrices. Therefore, it avoids the complexities and singularities associated with local coordinates of $\text{SO}(3)$, such as Euler angles. It also avoids the ambiguities that arise when using quaternions to represent the attitude dynamics. As the three-sphere S^3 double covers $\text{SO}(3)$, any attitude feedback controller designed in terms of quaternions could yield different

control inputs depending on the choice of quaternion vectors. The corresponding stability analysis would need to carefully consider the fact that convergence to a single attitude implies convergence to either of the two disconnected, antipodal points on S^3 [22]. This requires a continuous selection of the sign of quaternions or a discontinuous control system, which are shown to be sensitive to small measurement noise [23]. Without these considerations, a quaternion-based controller can exhibit an unwinding phenomenon, where the controller unnecessarily rotates the attitude through large angles [15]. In this paper, the use of rotation matrices in the controller design and stability analysis completely eliminates these difficulties.

Another novelty of the presented controller is the choice of the total thrust in (15). This is designed to follow position tracking commands, but it is also carefully designed to guarantee the overall stability of the complete dynamics by feedback control of the direction of the third body-fixed axis. This consideration is natural as each column of a rotation matrix represents the direction of each body-fixed axis. Therefore, another advantage of using rotation matrices is that the controller has a well-defined physical interpretation.

In Propositions 1 and 3, exponential stability and exponential attractiveness are guaranteed for almost all initial attitude errors, respectively. The attitude error function defined in (8) has the following critical points: the identity matrix, and rotation matrices that can be written as $\exp(\pi\hat{v})$ for any $v \in S^2$. These non-identity critical points of the attitude error function lie outside of the region of attraction. As it is a two-dimensional subspace of the three-dimensional $SO(3)$, we claim that the presented controller exhibits *almost global* properties in $SO(3)$. It is impossible to construct a smooth controller on $SO(3)$ that has global asymptotic stability. The two-dimensional family of non-identity critical points can be reduced to four points by modifying the error function to be $\frac{1}{2}\text{tr}[G(I - R_d^T R)]$ for a matrix $G \neq I \in \mathbb{R}^{3 \times 3}$. The presented controller can be modified accordingly.

IV. NUMERICAL EXAMPLE

The parameters of the quadrotor UAV are chosen according to a quadrotor UAV developed in [2].

$$J = [0.0820, 0.0845, 0.1377] \text{ kgm}^2, \quad m = 4.34 \text{ kg}$$

$$d = 0.315 \text{ m}, \quad c_{\tau f} = 8.004 \times 10^{-4} \text{ m.}$$

The controller parameters are chosen as follows:

$$k_x = 16m, \quad k_v = 5.6m, \quad k_R = 8.81, \quad k_\Omega = 2.54.$$

We consider the following two cases.

- (I) This maneuver follows an elliptical helix while rotating the heading direction at a fixed rate. Initial conditions are chosen as

$$x(0) = [0, 0, 0], \quad v(0) = [0, 0, 0],$$

$$R(0) = I, \quad \Omega(0) = [0, 0, 0].$$

The desired trajectory is as follows.

$$x_d(t) = [0.4t, 0.4 \sin \pi t, 0.6 \cos \pi t],$$

$$\vec{b}_{1_d}(t) = [\cos \pi t, \sin \pi t, 0].$$

- (II) This maneuver recovers from being initially upside down. Initial conditions are chosen as

$$x(0) = [0, 0, 0], \quad v(0) = [0, 0, 0],$$

$$R(0) = \begin{bmatrix} 1 & 0 & 0 \\ 0 & -0.9995 & -0.0314 \\ 0 & 0.0314 & -0.9995 \end{bmatrix}, \quad \Omega(0) = [0, 0, 0].$$

The desired trajectory is as follows.

$$x_d(t) = [0, 0, 0], \quad \vec{b}_{1_d}(t) = [1, 0, 0].$$

Simulation results are presented in Figures 4 and 5. For Case (I), the initial value of the attitude error function $\Psi(0)$ is less than 0.15. This satisfies the conditions for Proposition 2, and exponential asymptotic stability is guaranteed. As shown in Figure 4, the tracking errors exponentially converge to zero. This example illustrates that the proposed controlled quadrotor UAV can follow a complex trajectory that involve large angle rotations and nontrivial translations accurately.

In Case (II), the initial attitude error is 178° , which yields the initial attitude error function $\Psi(0) = 1.995 > 1$. This corresponds to Proposition 3, which implies almost global exponential attractiveness. In Figure 5(b), the attitude error function Ψ decreases, and it becomes less than 1 at $t = 0.88$ seconds. After that instant, the position tracking error and the angular velocity error converge to zero as shown in Figures 5(c) and 5(d). The region of attraction of the proposed control system almost covers $SO(3)$, so that the corresponding controlled quadrotor UAV can recover from being initially upside down.

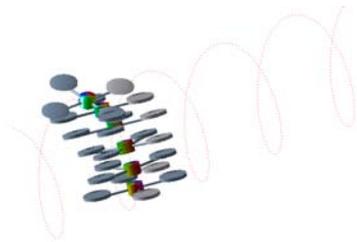
V. CONCLUSION

We presented a global dynamic model for a quadrotor UAV, and we developed a geometric tracking controller directly on the special Euclidean group that is intrinsic and coordinate-free, thereby avoiding the singularities of Euler angles and the ambiguities of quaternions in representing attitude. It exhibits exponential stability when the initial attitude error is less than 90° , and it yields almost global exponential attractiveness when the initial attitude error is less than 180° . These are illustrated by numerical examples.

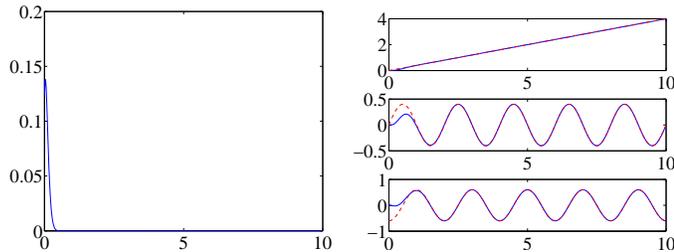
This controller can be extended as follows. In this paper, four input degrees of freedom are used to track a three-dimensional position, and a one-dimensional heading direction. But, without changing the controller structure, they can be used to follow arbitrary three-dimensional attitude commands. The remaining one input degree of freedom can be used to maintain the altitude as much as possible. By constructing a hybrid controller based on these two tracking modes, we can generate complicated acrobatic maneuvers of a quadrotor UAV.

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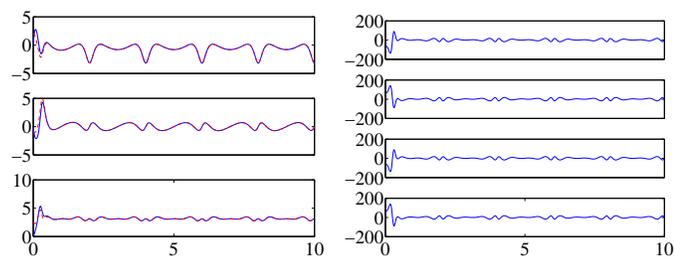
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(a) Snapshots for $2 \leq t \leq 2.6$ (an animation illustrating this maneuver is available at <http://my.fit.edu/~taeyoung>)



(b) Attitude error function Ψ (c) Position (x :solid, x_d :dotted, (m))

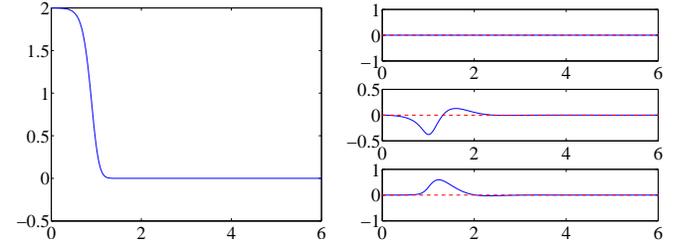


(d) Angular velocity (Ω :solid, Ω_d :dotted, (rad/sec)) (e) Thrust of each rotor (N)

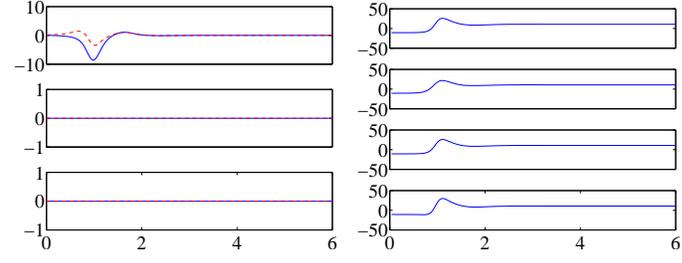
Fig. 4. Case I: following an elliptic helix (horizontal axes represent simulation time in seconds)



(a) Snapshots for $0.5 \leq t \leq 4$ (Snapshots are shifted forward to represent the evolution of time. In reality, the quadrotor is flipped at a fixed position. An animation is available at <http://my.fit.edu/~taeyoung>)



(b) Attitude error function Ψ (c) Position (x :solid, x_d :dotted, (m))



(d) Angular velocity (Ω :solid, Ω_d :dotted, (rad/sec)) (e) Thrust of each rotor (N)

Fig. 5. Case II: recovering from an initially upside down attitude (horizontal axes represent simulation time in seconds)

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